



SUBJECT: SPIRE Failure Detection Isolation and Recovery

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Project Document

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and Recovery**

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Change Record

ISSUE	DATE	Changes
1.0 Draft 1	17 th March 2004	First draft for internal discussion
1.0 Draft 2	16 th June 2004	Updated draft for internal review
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Glossary

OBS On-Board Software
SPIRE Spectral and Photometric Imaging REceiver



1. INTRODUCTION

1.1 Scope

This document defines the requirements on the SPIRE instrument for detection, isolation and recovery from hardware and software failures of the instrument based on the policy described in AD03. The failures to be dealt with are described in AD01 and AD02.

These requirements will lead to requirements on the OBS autonomy function to detect and isolate the failure and provide a mechanism for recovery from it. The implementation of the recovery procedures will be made in the OBS itself and/or in recovery procedures defined for the CDMS.

In this document FDIR actions are described in flowchart form. Actions carried out by the OBS in internal code are labelled 'OBS' and those carried out by the Autonomy Virtual Machine are labelled 'VM'. Actions carried out by the CDMS are labelled 'S/C'.

1.2 Assumptions

1. 'Inhibit TC Execution' implies that the instrument will no longer execute any telecommands received. It will, however, respond with a telecommand reception acknowledgement. This is to stop any newly received TC restarting a VM or sending a command to a subsystem while the recovery is in progress.
2. 'Inhibit Subsystem Commands' implies that all telecommands received by the OBS which would generate commands to a subsystem are inhibited. The OBS is still able to send commands to the subsystem as part of the recovery procedure.
3. The FDIR is suspended on the first instance of an anomaly being detected. I.e. the Autonomy Virtual Machine cannot be interrupted by another error.

1.3 Structure of Document

The failures themselves can be split into two types: those detected by the instrument OBS and handled by it (possibly with help by the S/C CDMS) – these are described in Section 2 (subsystem failures), Section 3 (interface failures), and Section 4 (OBS runtime errors); and those detected by the S/C CDMS and dealt with by it – these are described in section 5.

Section 6 contains those general purpose procedures called during some of the recovery actions described in previous sections

1.4 Documents

1.4.1 Applicable Documents

- AD01 Hardware Software Interaction Analysis for SPIRE in-flight Autonomy Functions Specification (SPIRE-RAL-NOT-001719), Issue 1.1, 3rd December 2003
- AD02 System Operation and FDIR Requirements (H-P-1-ASP-SP-0209), October 2003
Appendix 1: 1553 Bus FDIR
- AD03 Failure Detection Isolation and Recovery Policy in the SPIRE Instrument (SPIRE-RAL-PRJ-001128)

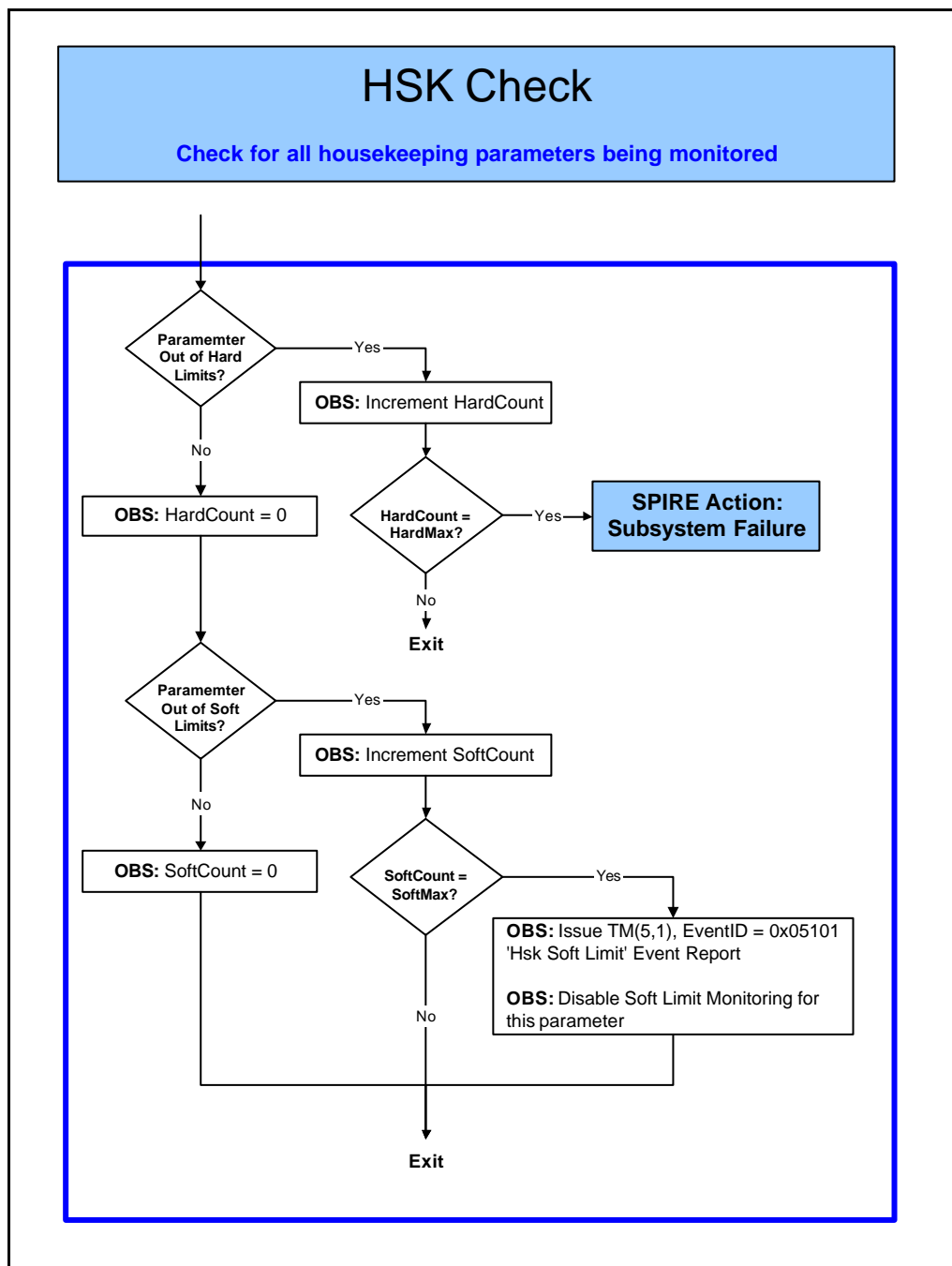
1.4.2 Reference Documents



2. SUBSYSTEM FAILURES

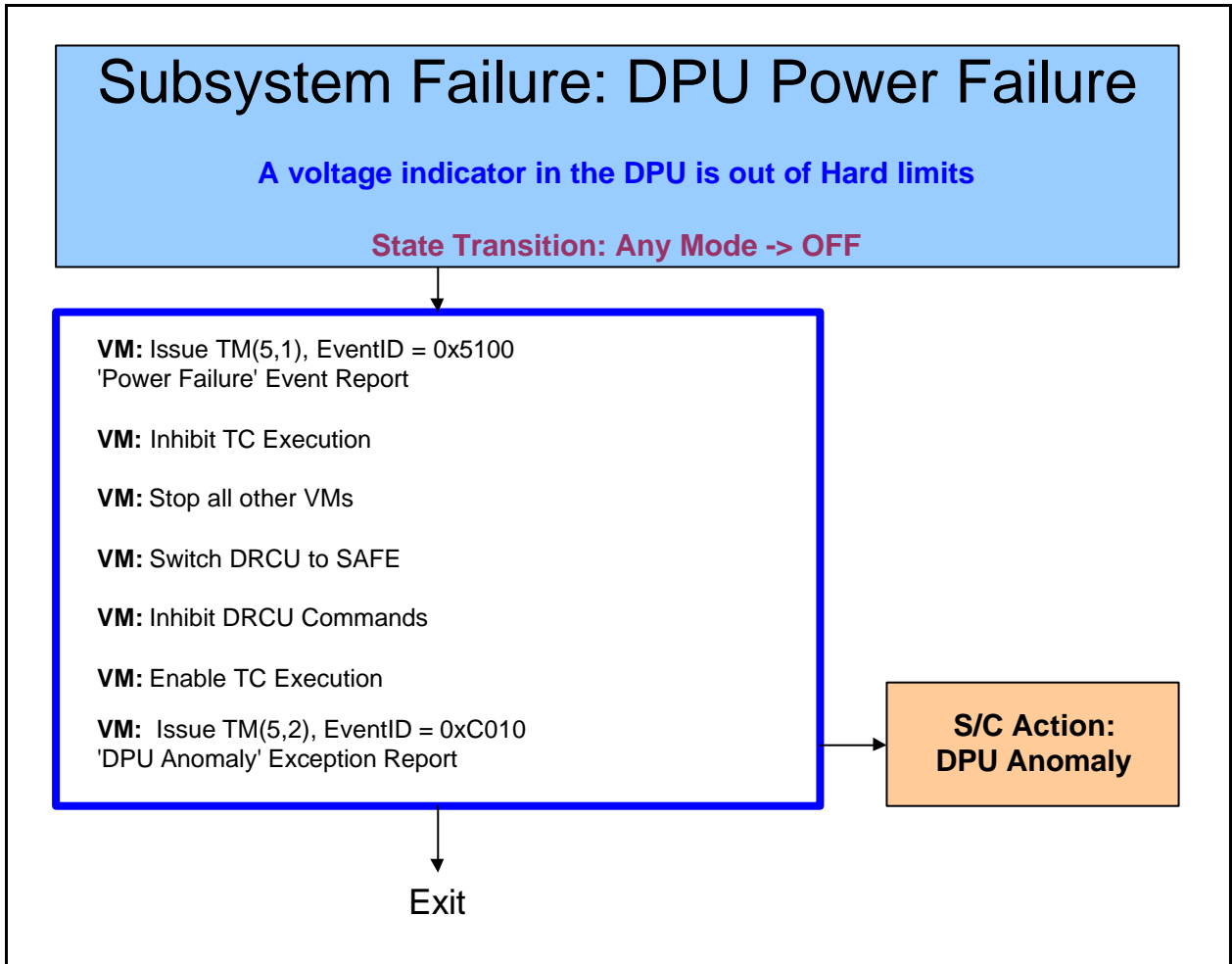
Failures within the subsystems of the instrument (both DPU and DRCU) may be identified by monitoring housekeeping parameters generated by the subsystem. Each housekeeping parameter being monitored is checked against a set of soft and hard limits. The parameter has to remain out of limits for a set number (SoftMax or HardMax) of readings before an anomaly is declared. At the soft limit monitoring is disabled once reported, to prevent multiple event packets being generated from a 'noisy' signal. It is expected that it will be re-enabled at the start of each observation, or other suitable time period.

SoftMax and HardMax are set to zero on initialisation of the parameter monitoring system



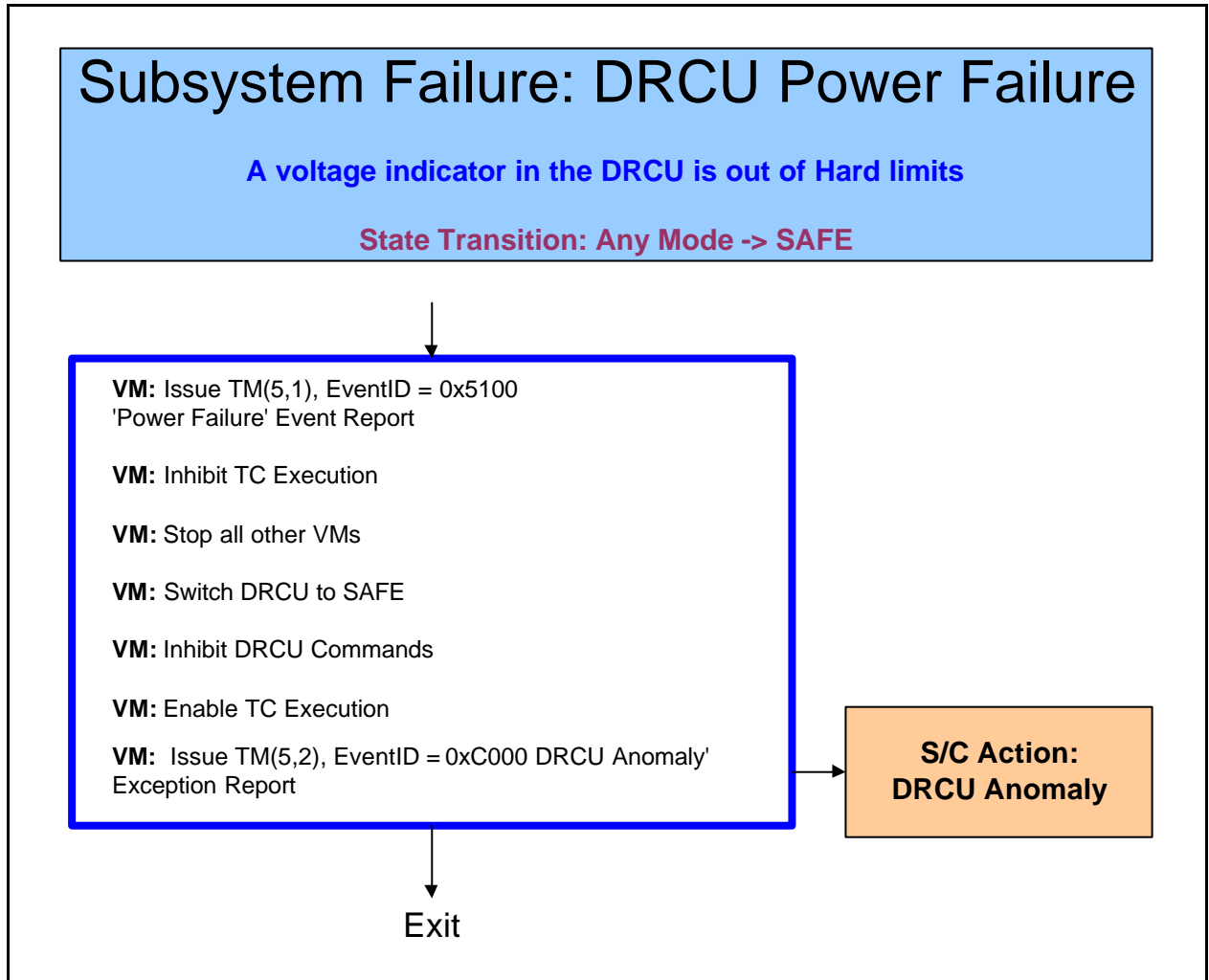


2.1 DPU Power

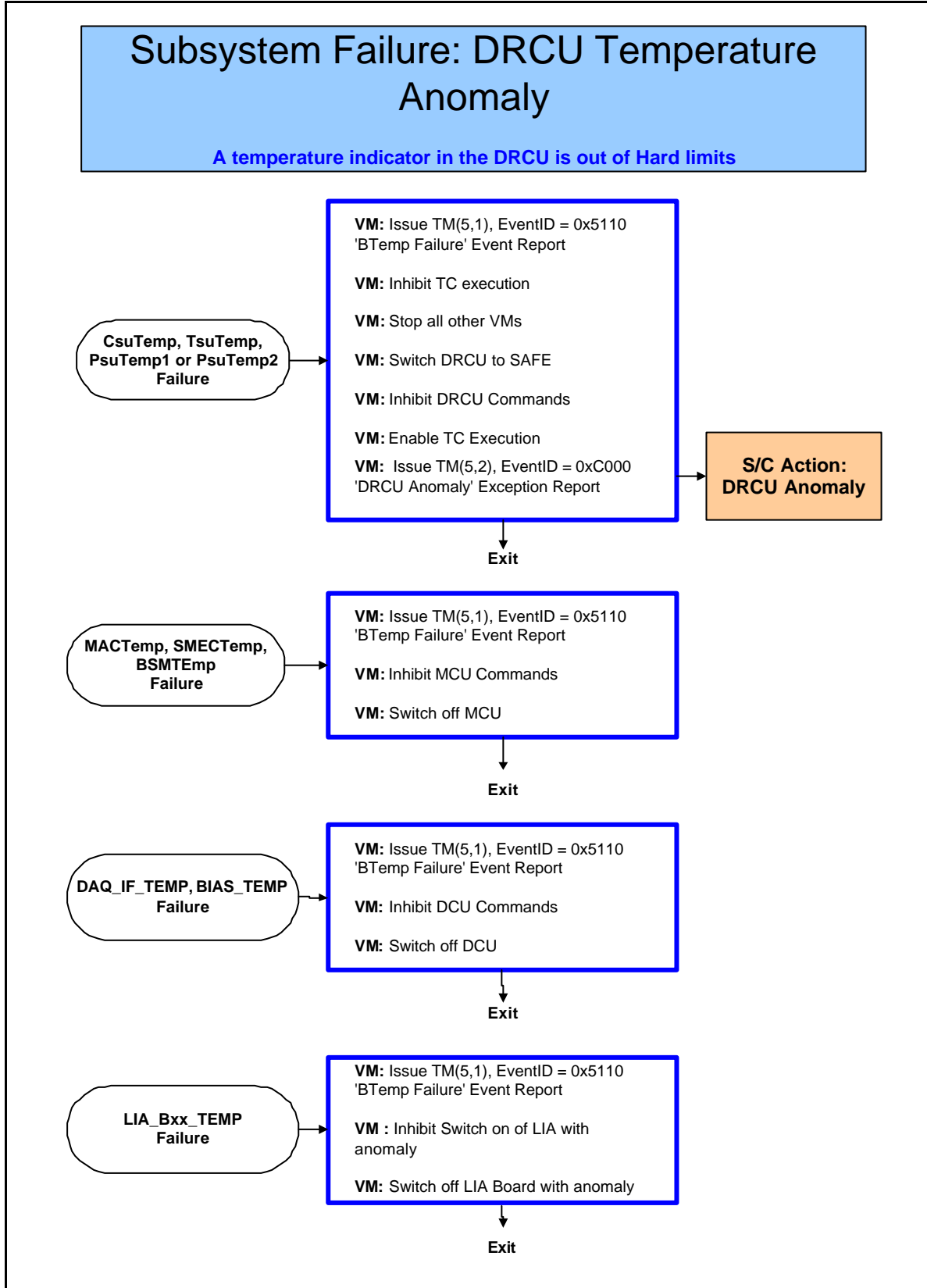




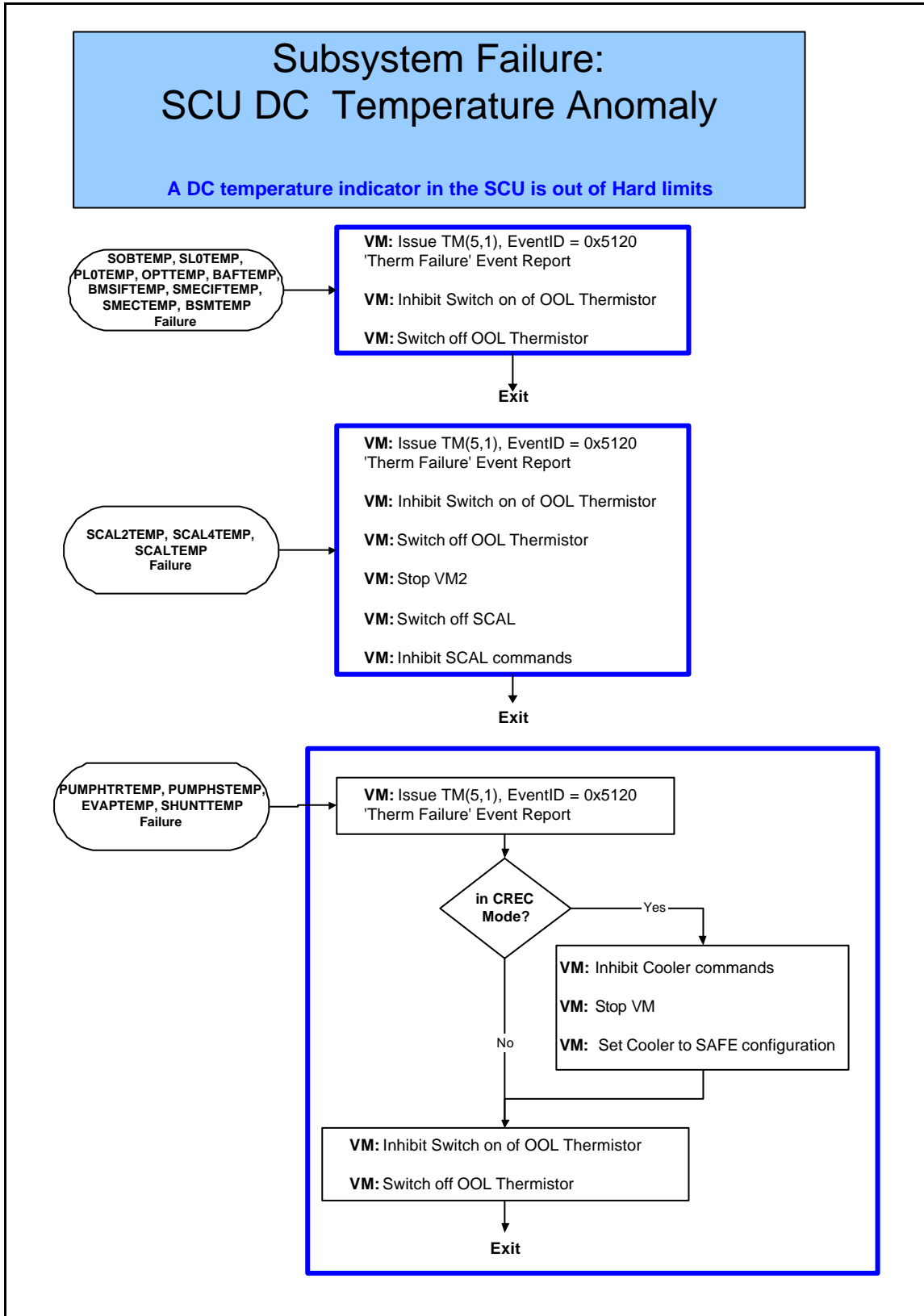
2.2 DRCU Power



2.3 DRCU Board Temperatures

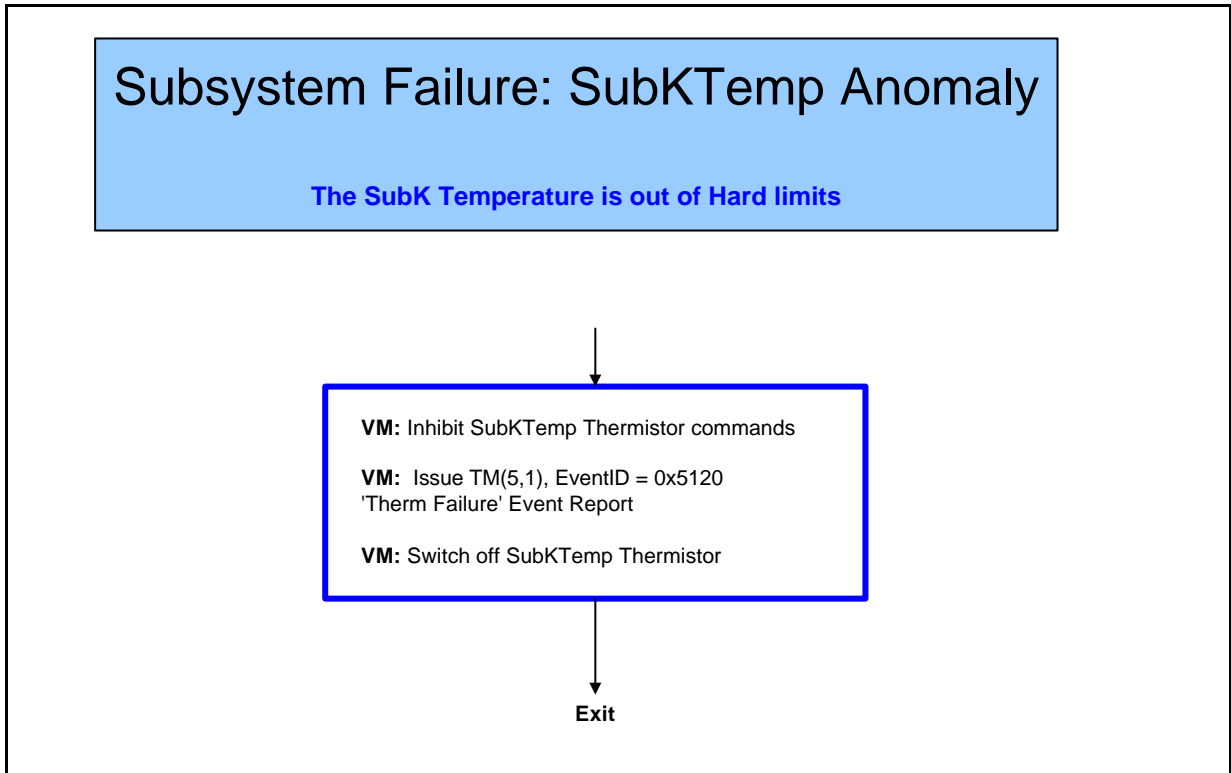


2.4 DC Thermistors

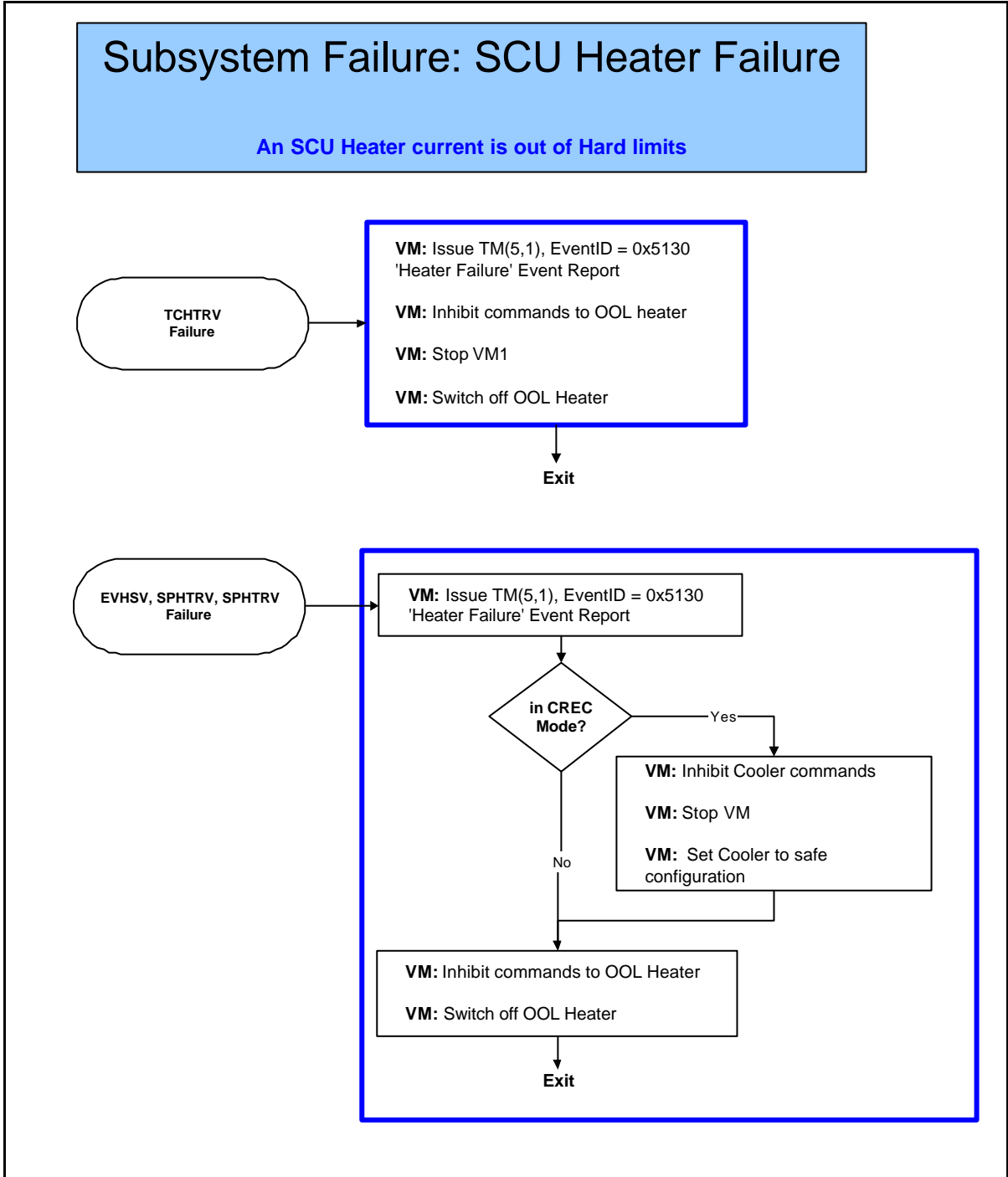




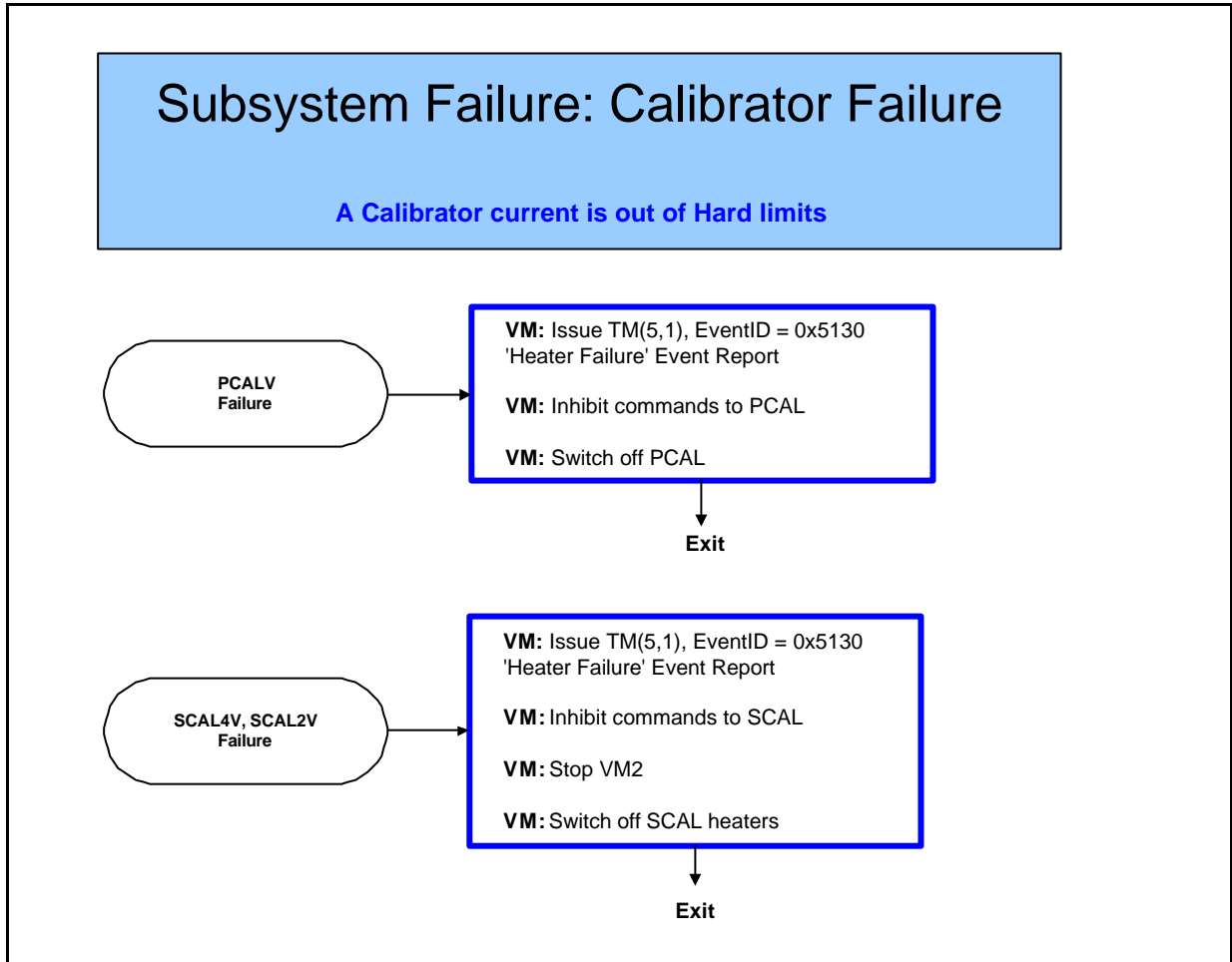
2.5 AC Thermistor



2.6 Heaters

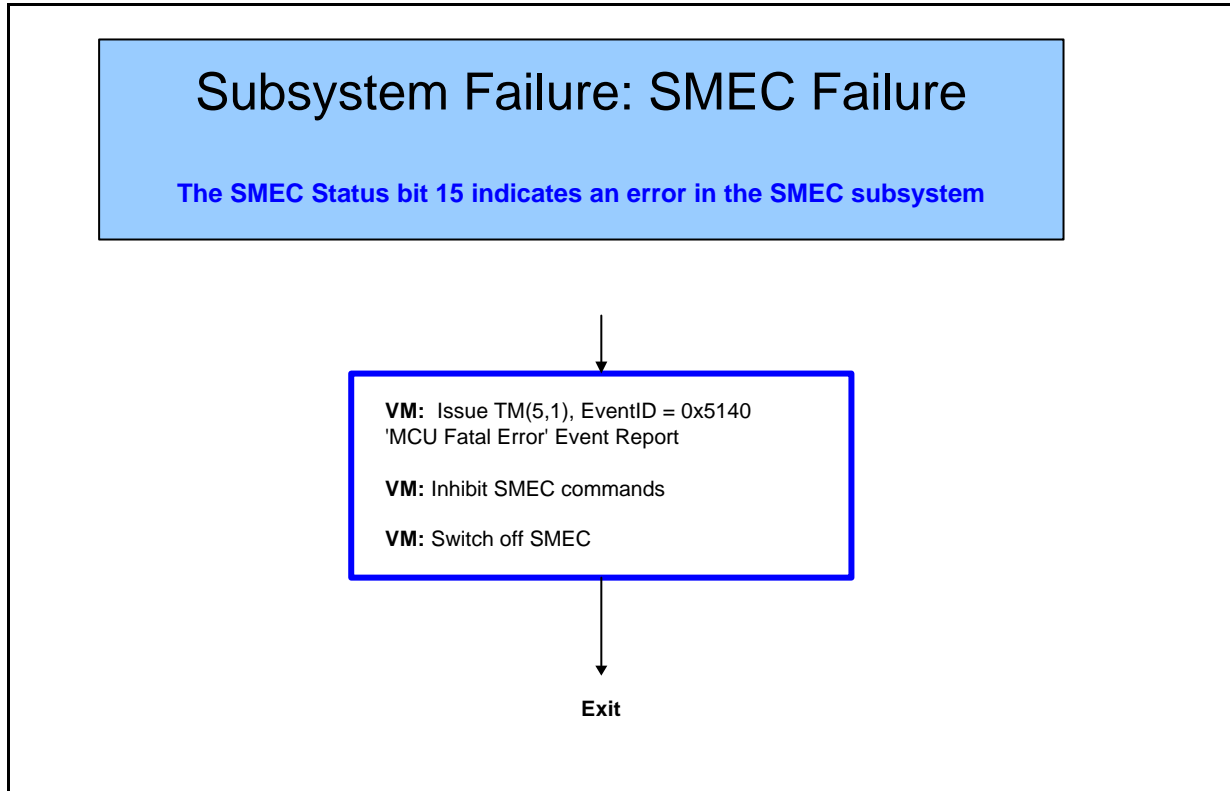


2.7 Calibrators

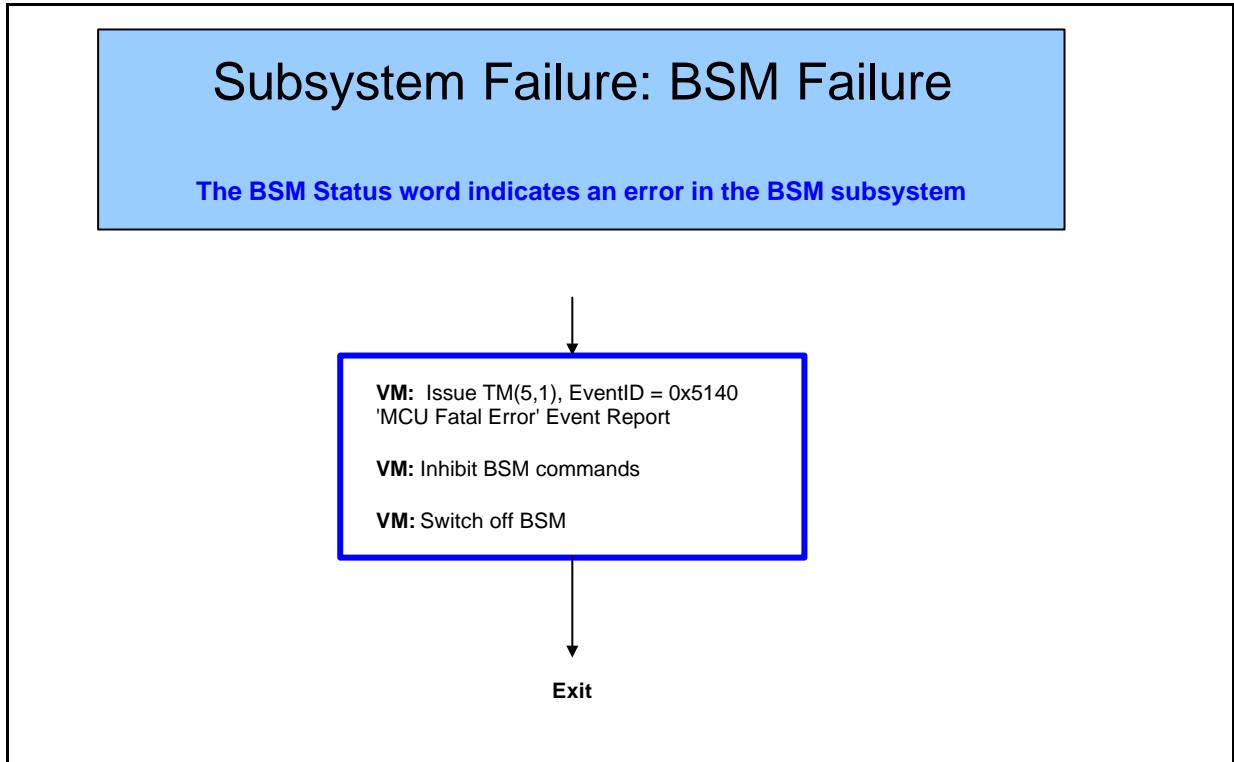




2.8 SMEC

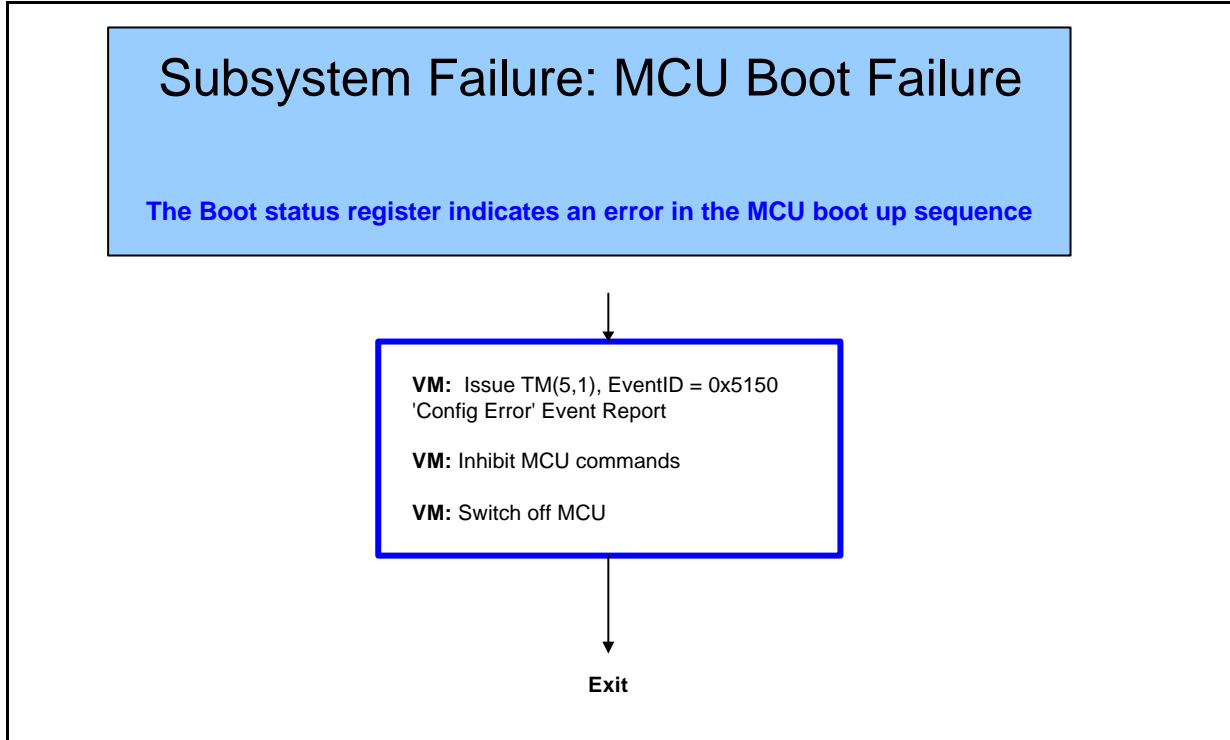


2.9 BSM





2.10 MCU DSP





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2.11 DCU

No specific DCU monitoring has been identified



3. DRCU INTERFACES

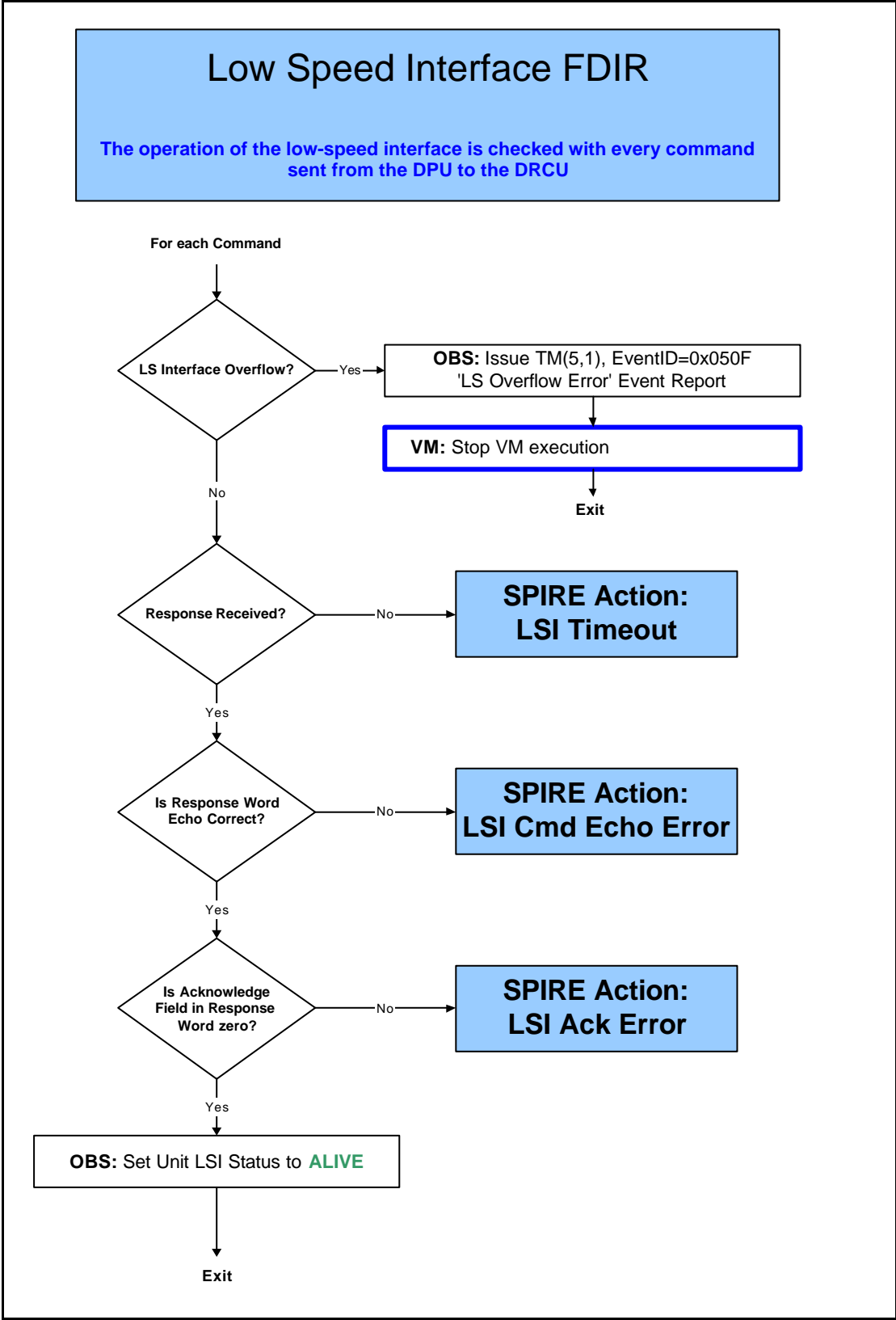
Subsystems are commanded and monitored through data passing through the interfaces with the DRCU. These interfaces have to be checked to be operating correctly before correct operation of the instrument can be expected.

The DRCU contains three units; the DCU, MCU and SCU, and each one has one slow speed interface (used to send commands to the unit and to collect housekeeping parameters) and one high speed interface (used to transfer science data from the unit to the DPU).

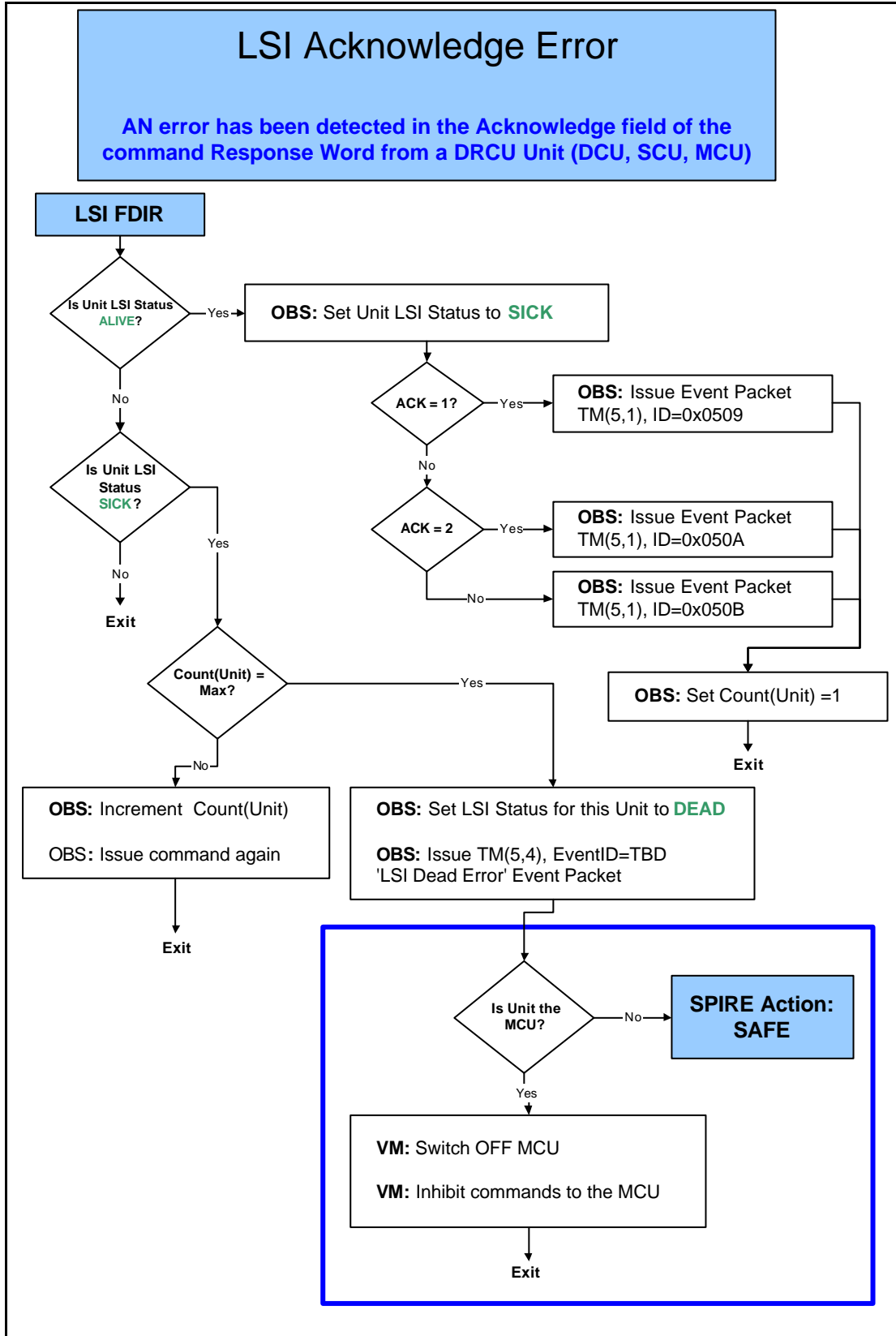
3.1 Low Speed Interface Failures

For each command sent through the LSI a series of checks is made to ensure that the command has been executed correctly. In the event of failure the command is repeated a set number (LSIMax) of times before a failure is declared.

RespMax is set to zero on initialisation of the autonomy system



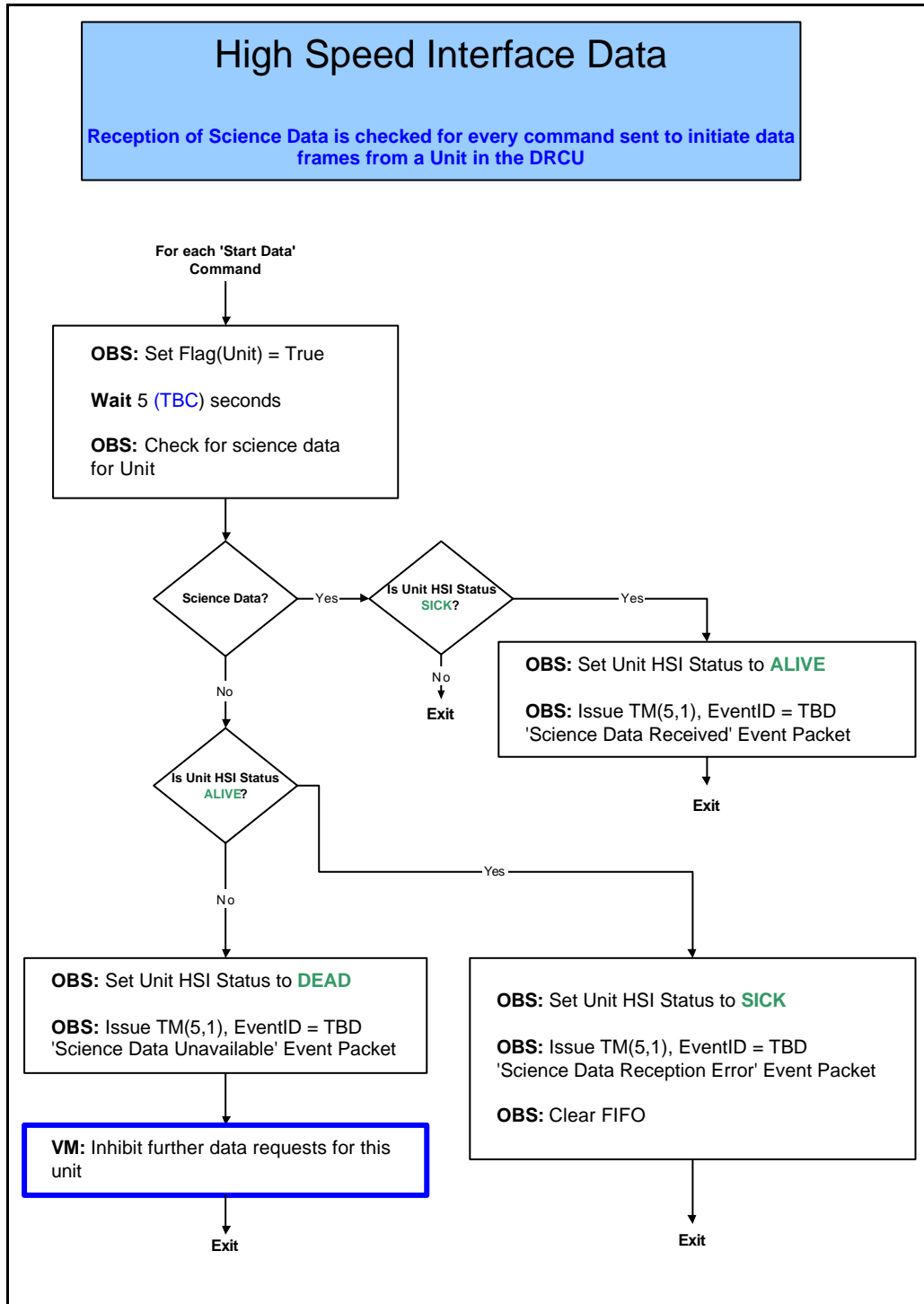
3.1.3 LSI Acknowledge



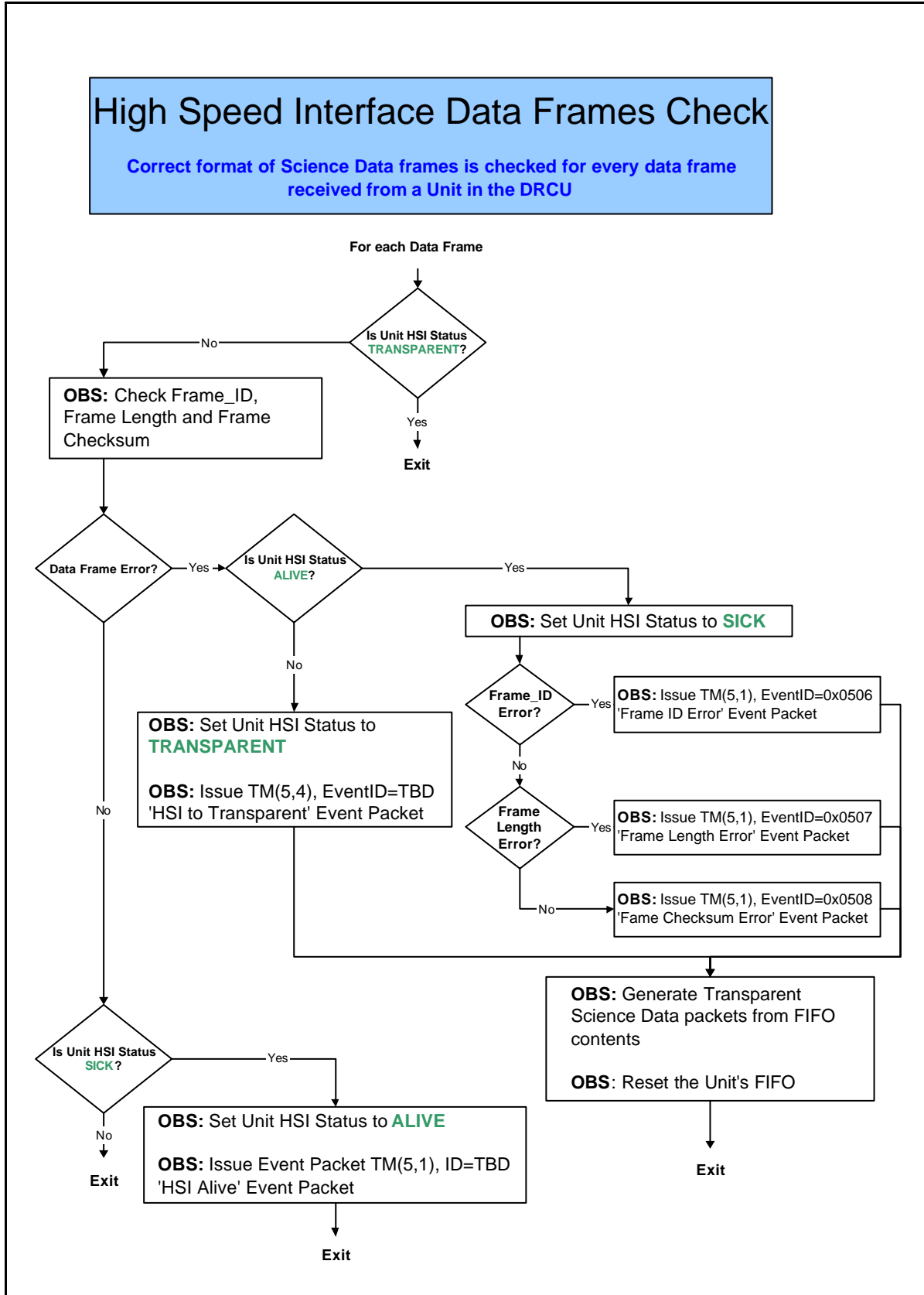
3.2 High Speed Interface Failures

The high speed interfaces transfer science data from the DRCU units to the DPU. Data is collected by the DPU into FIFO buffers which trigger the DPU to empty them when they are half full.

3.2.1 HSI Data

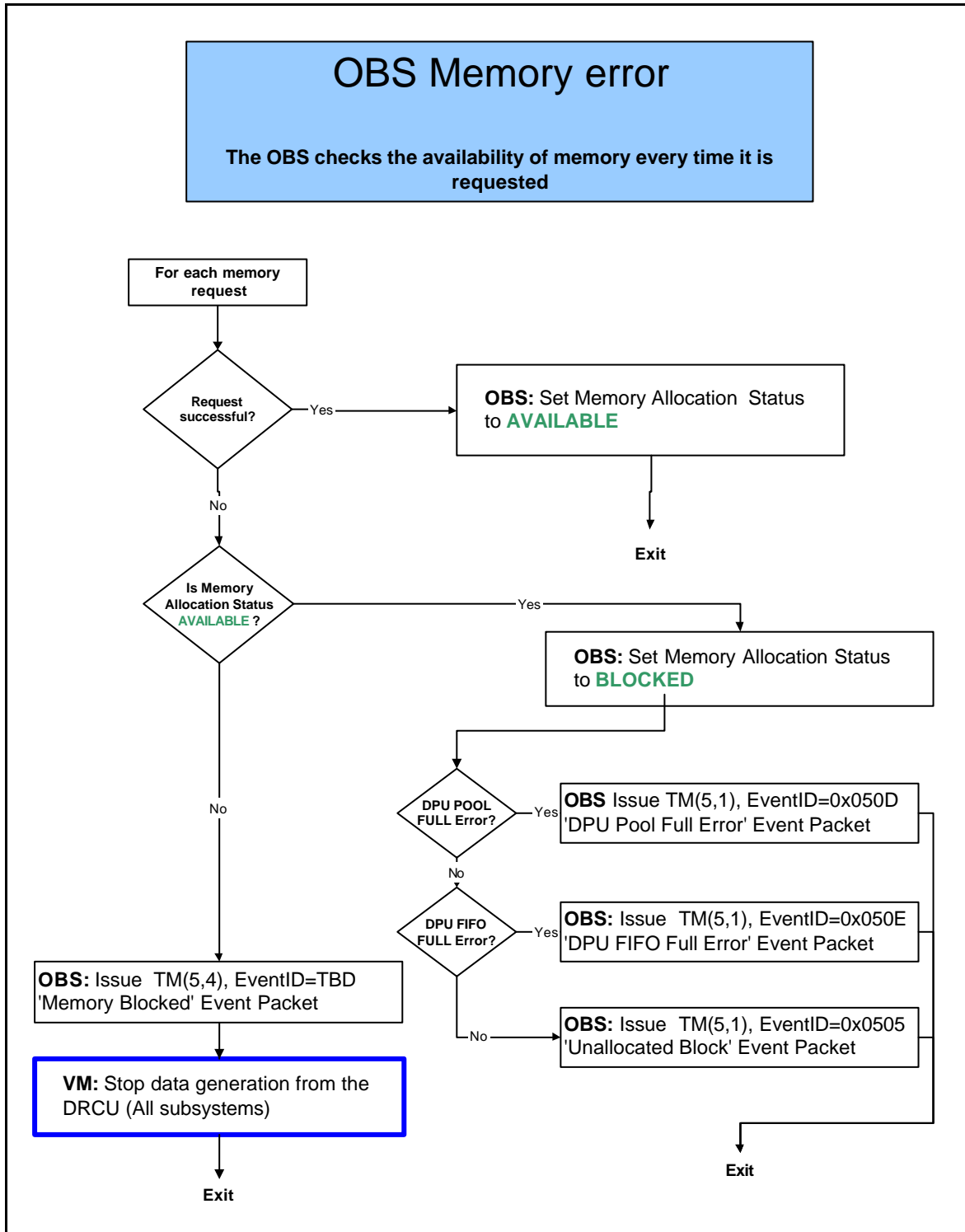


3.2.2 HSI Frames



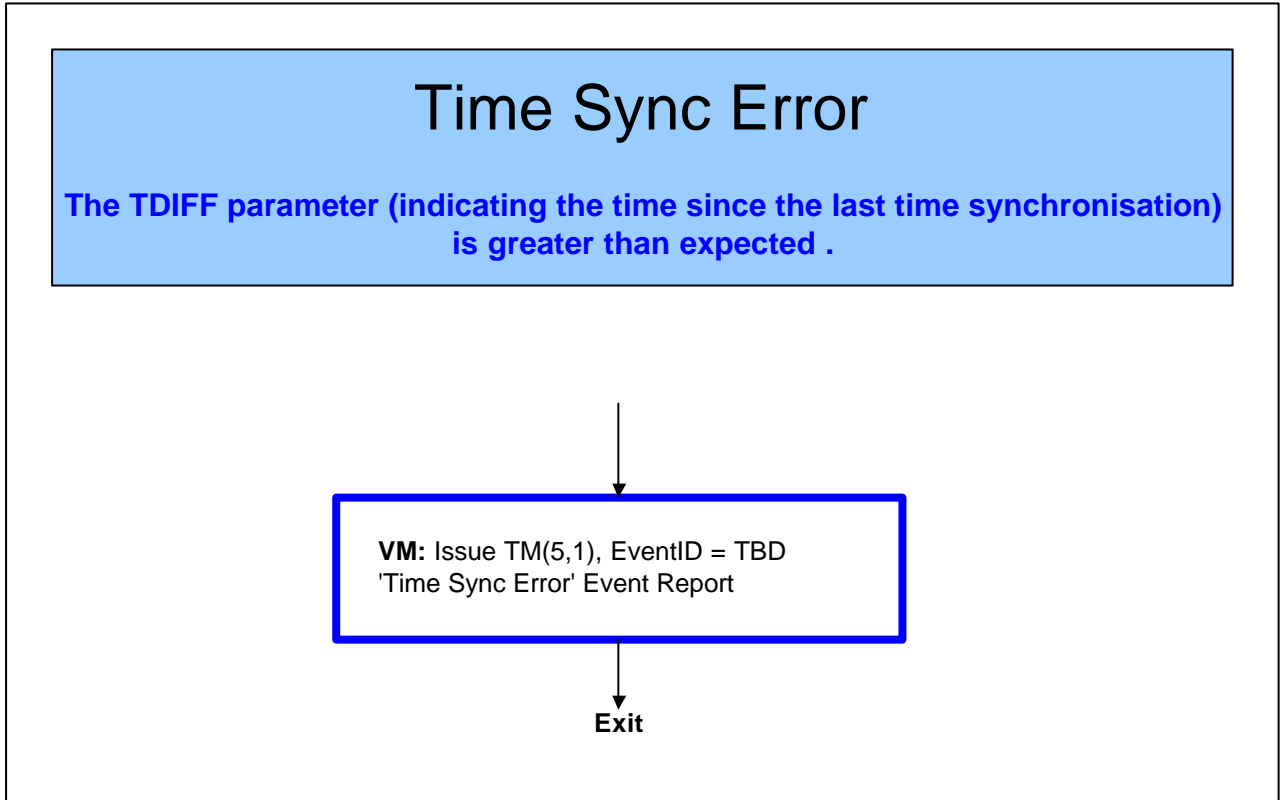
4. OBS RUNTIME ERRORS

4.1 Memory Errors





4.2 Time Sync Error



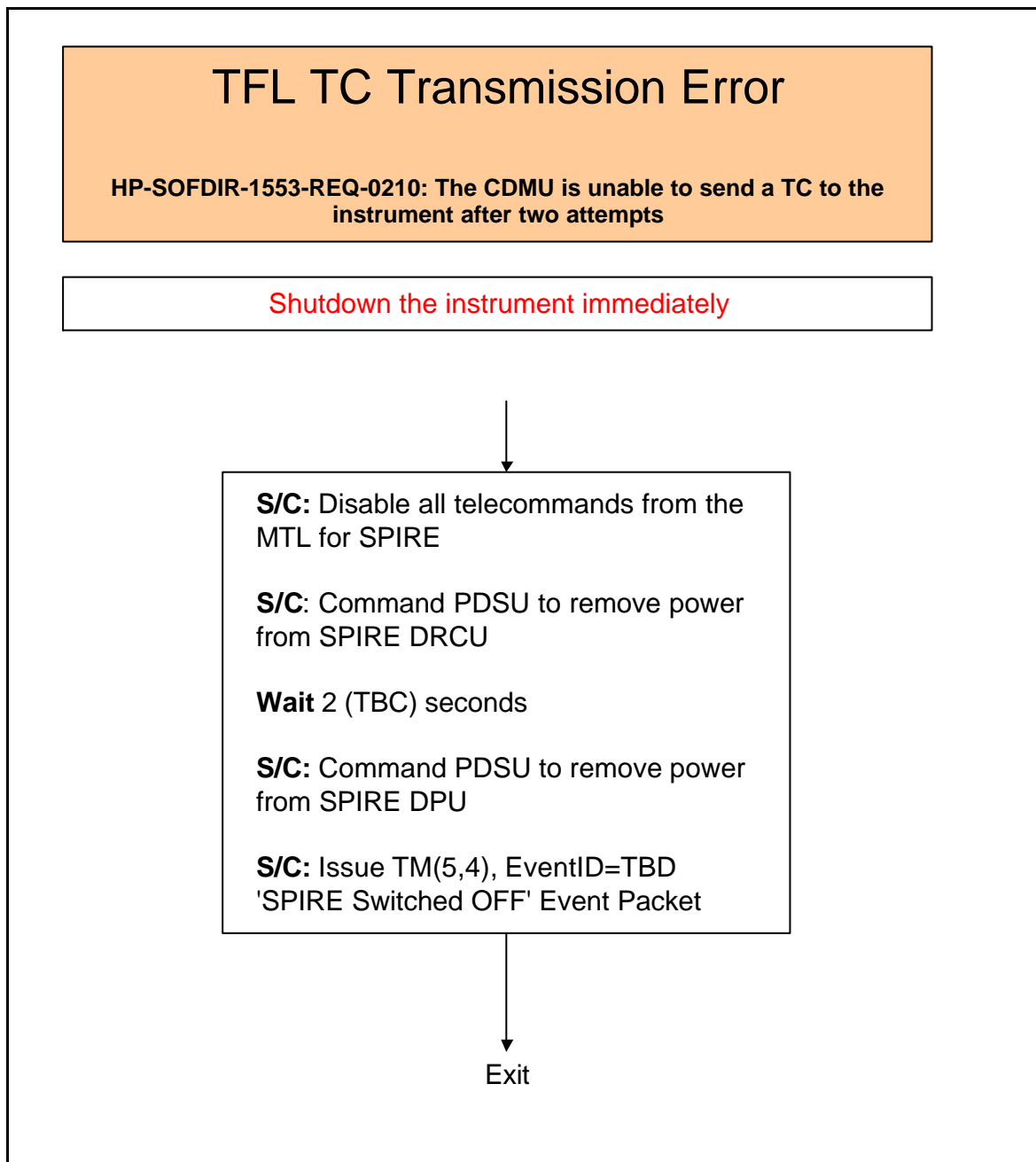


5. S/C INTERFACE

5.1 Bus Failures

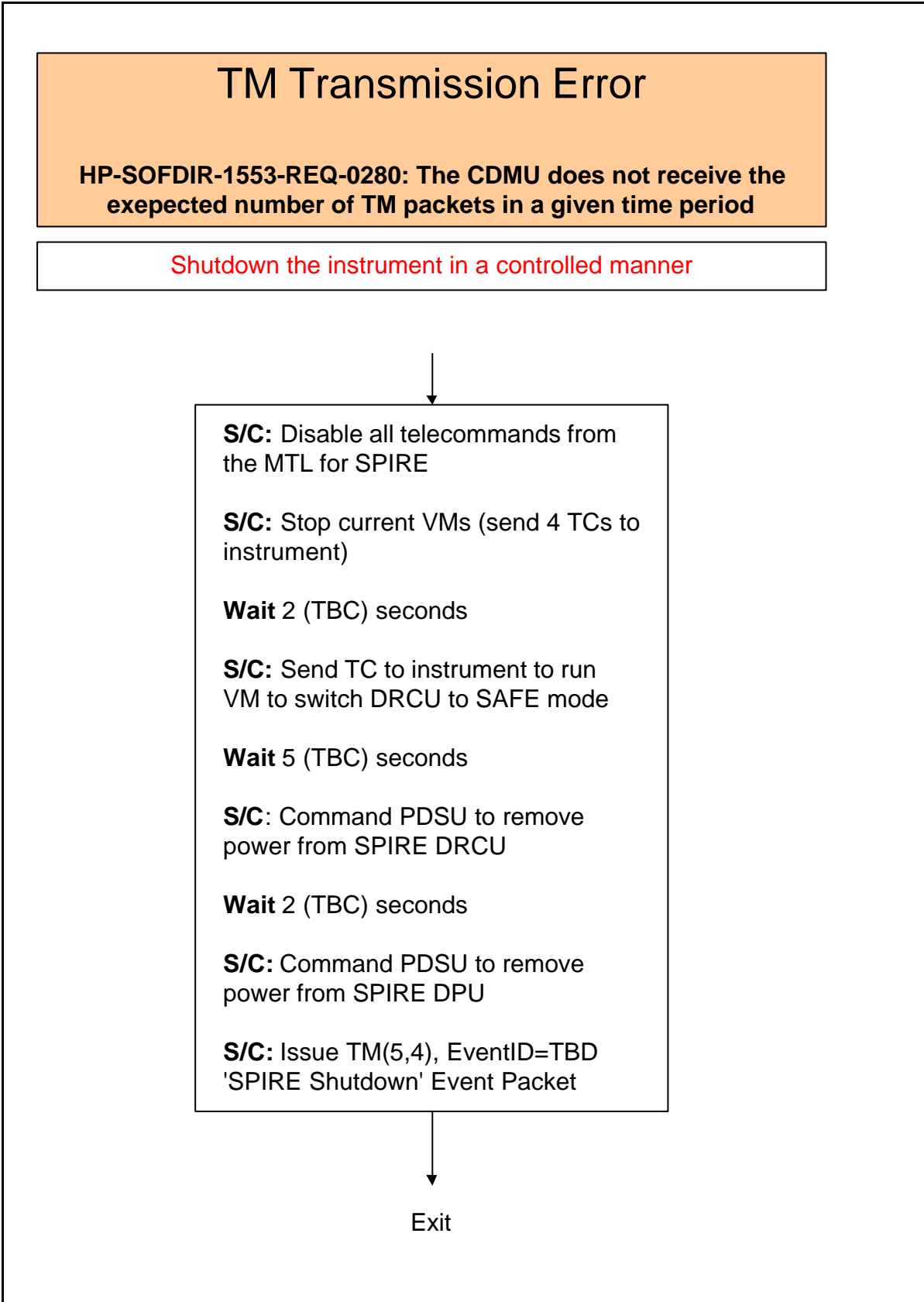
Detected by the S/C SOFDIR

5.1.1 TC Transmission

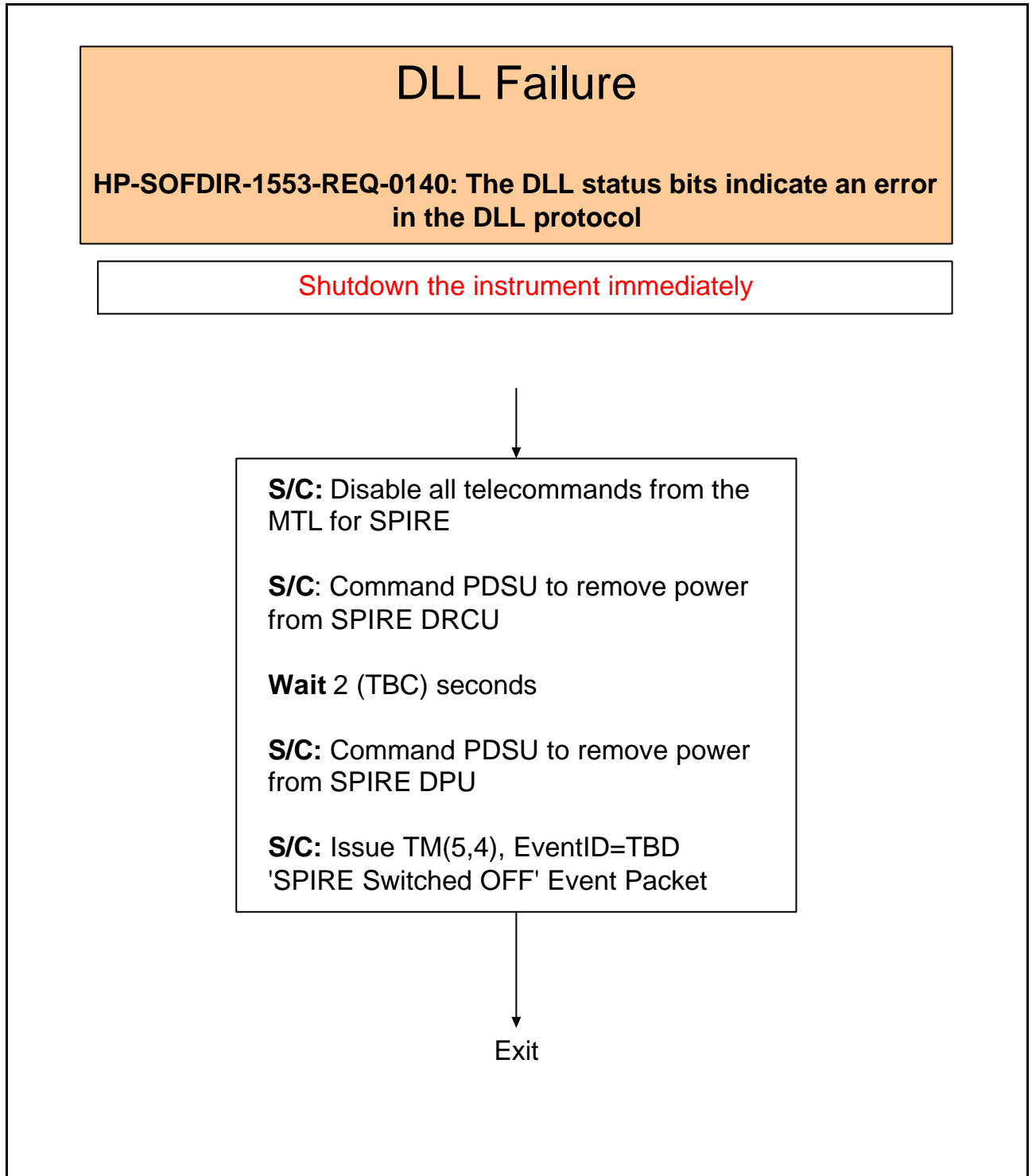




5.1.2 TM Transmission

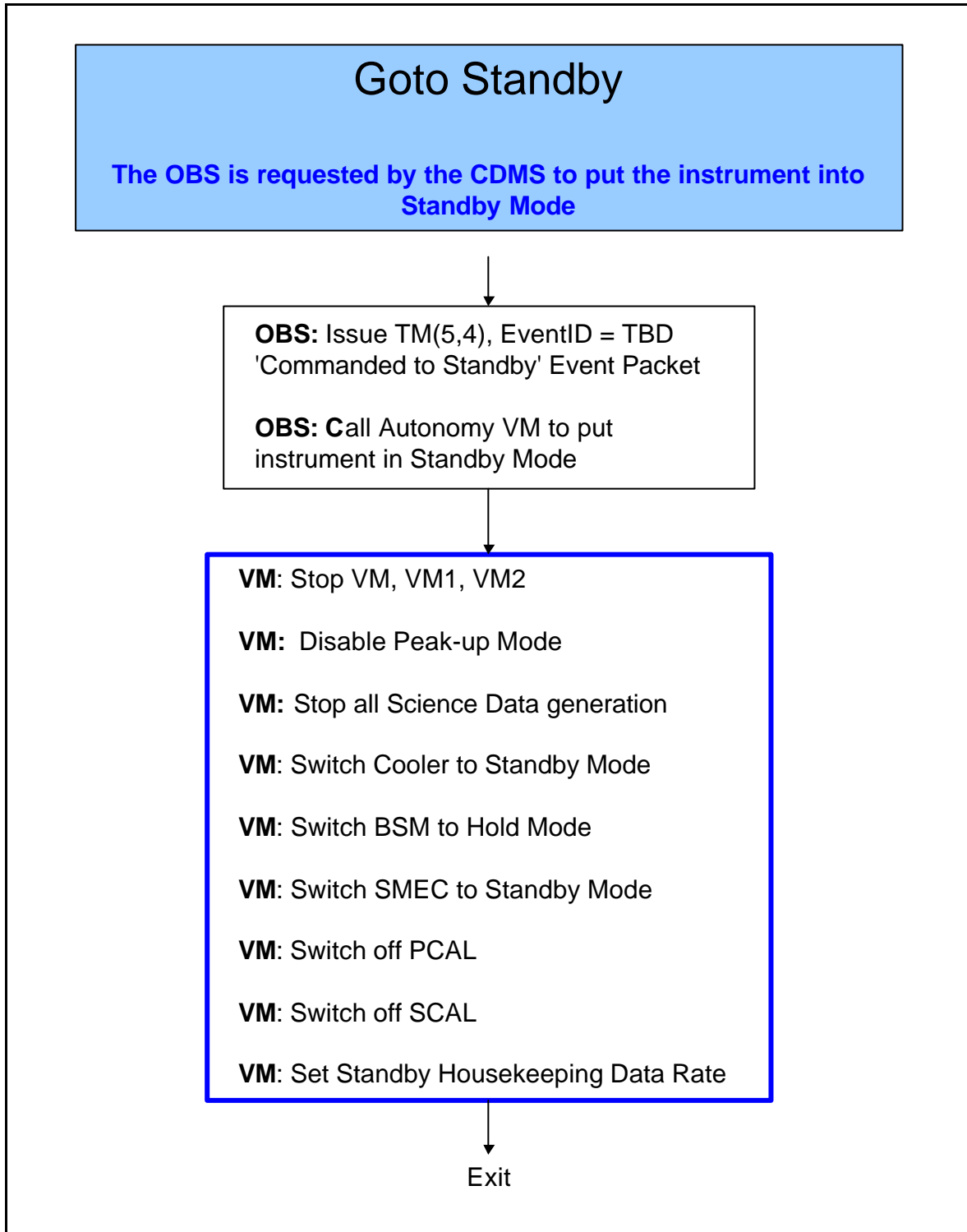


5.1.3 DLL



5.2 Spacecraft Commands

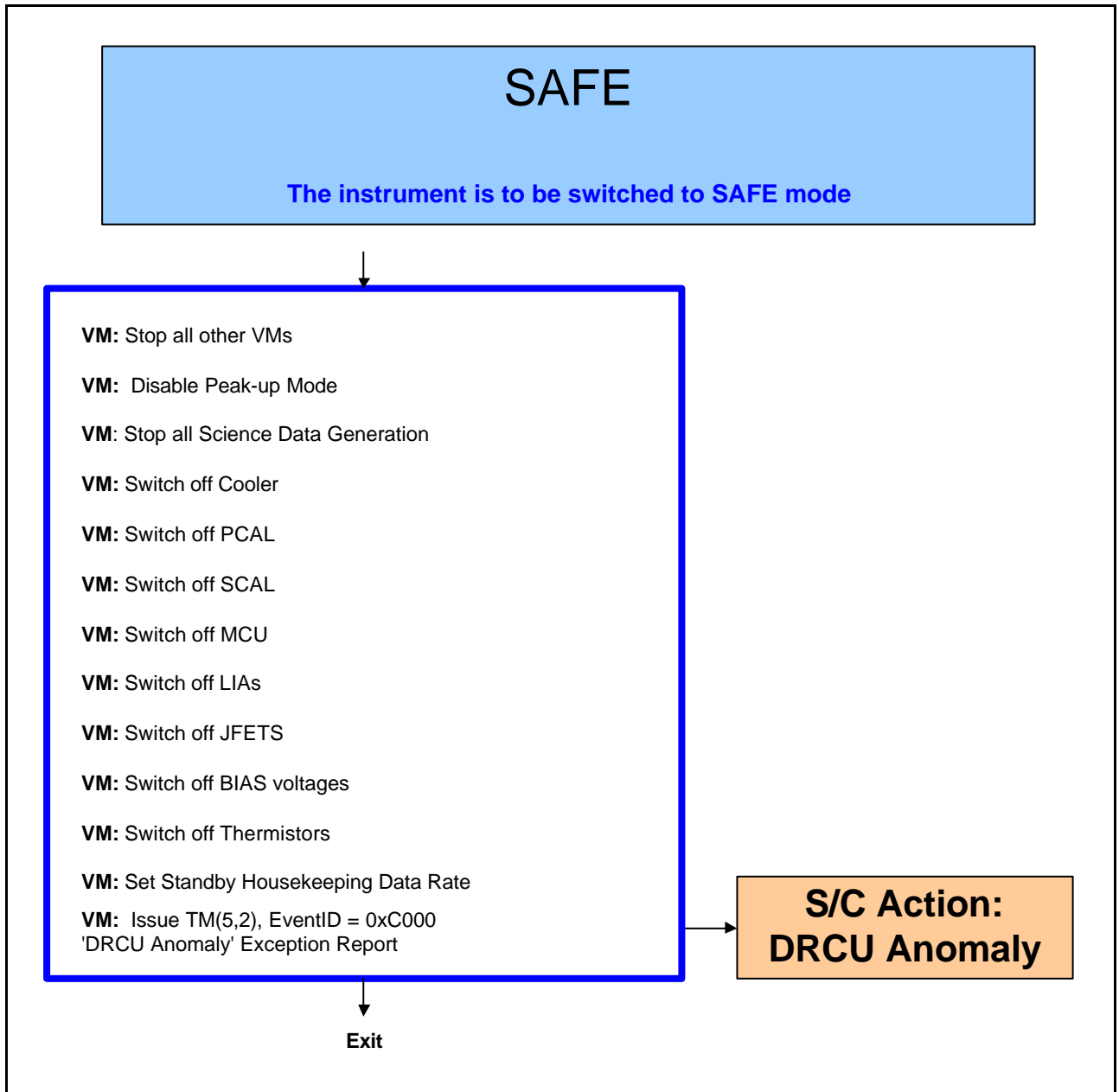
5.2.1 Go to Standby





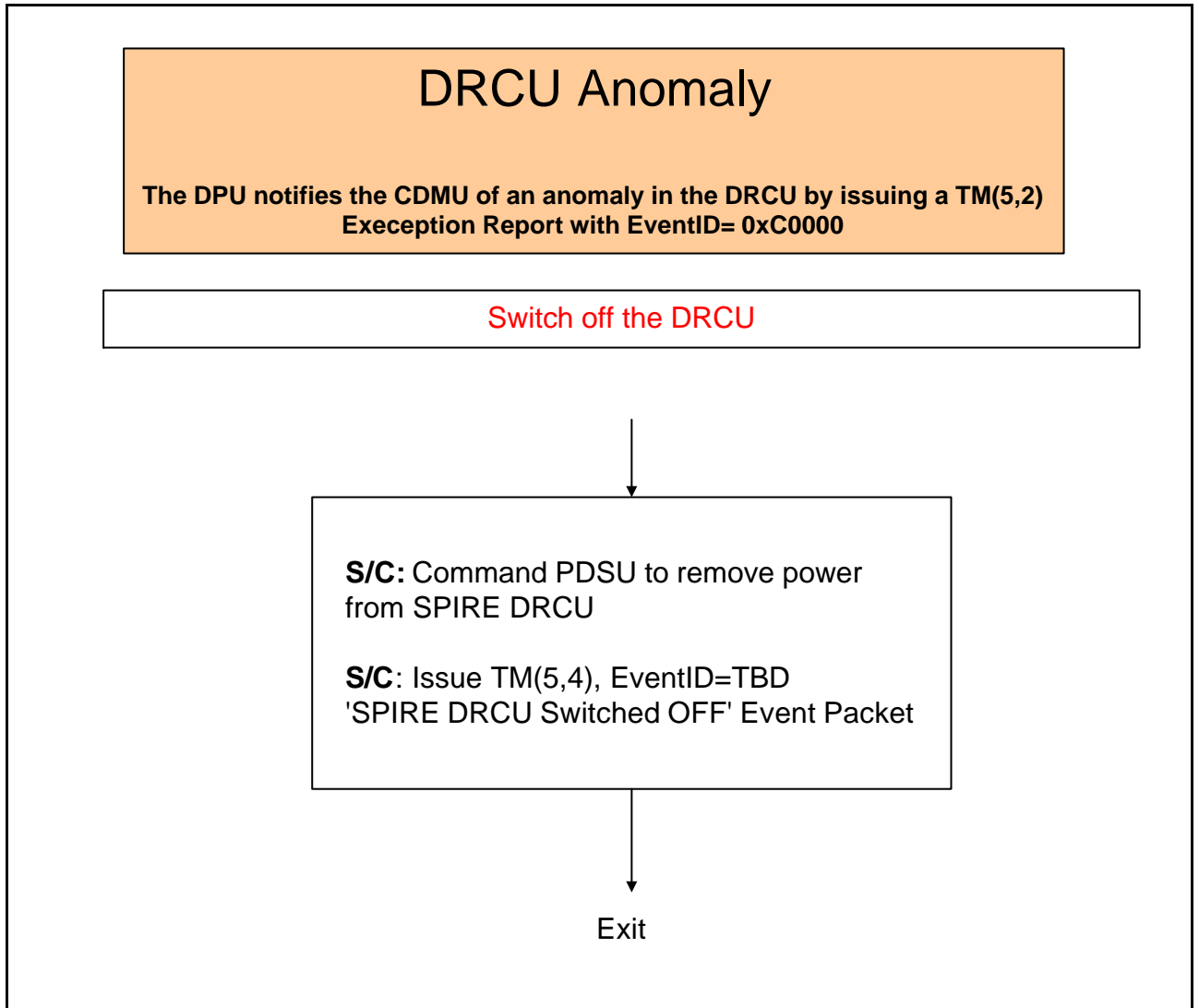
6. GENERAL PURPOSE PROCEDURES

6.1 SAFE



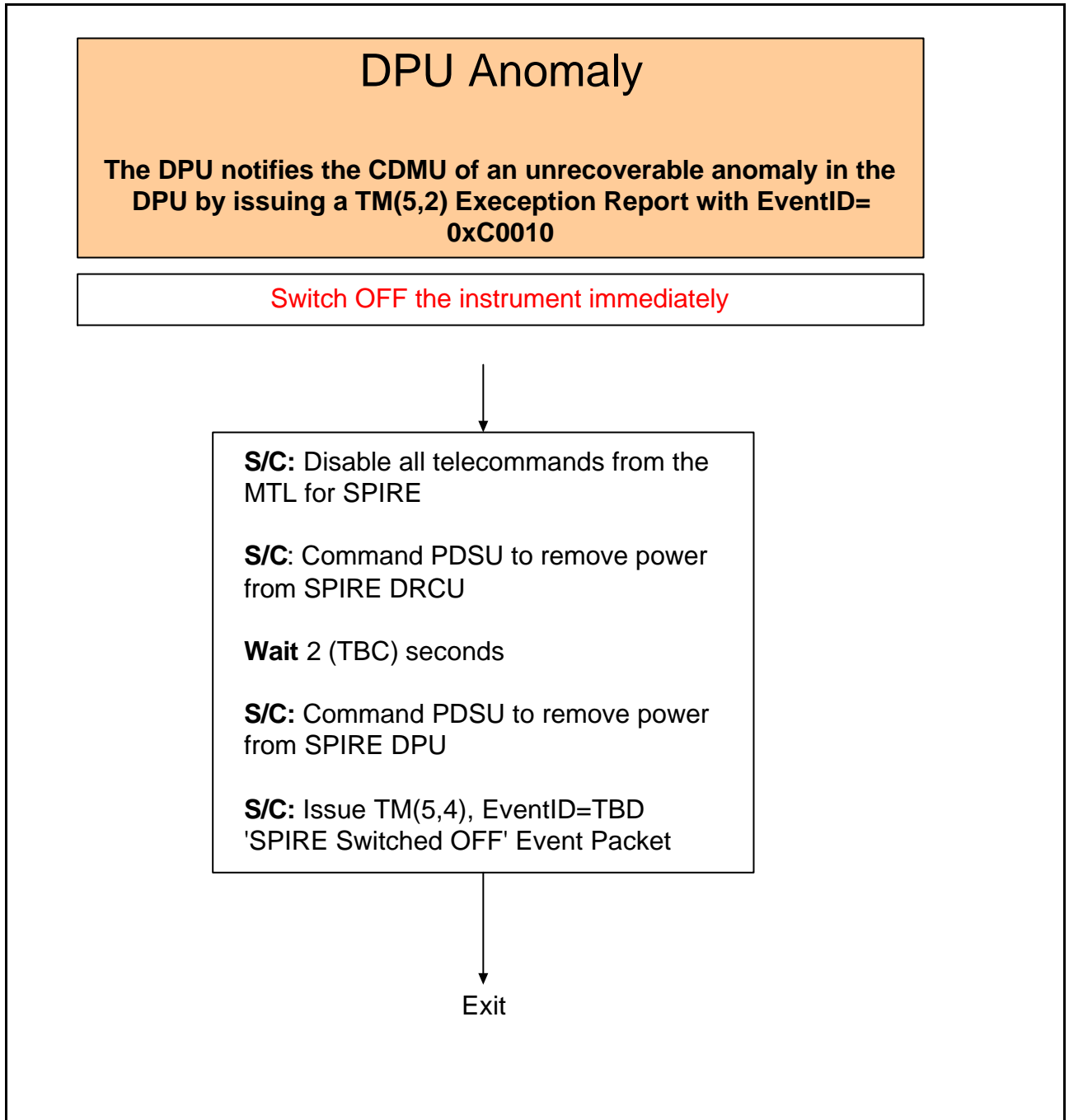


6.2 DRCU Anomaly



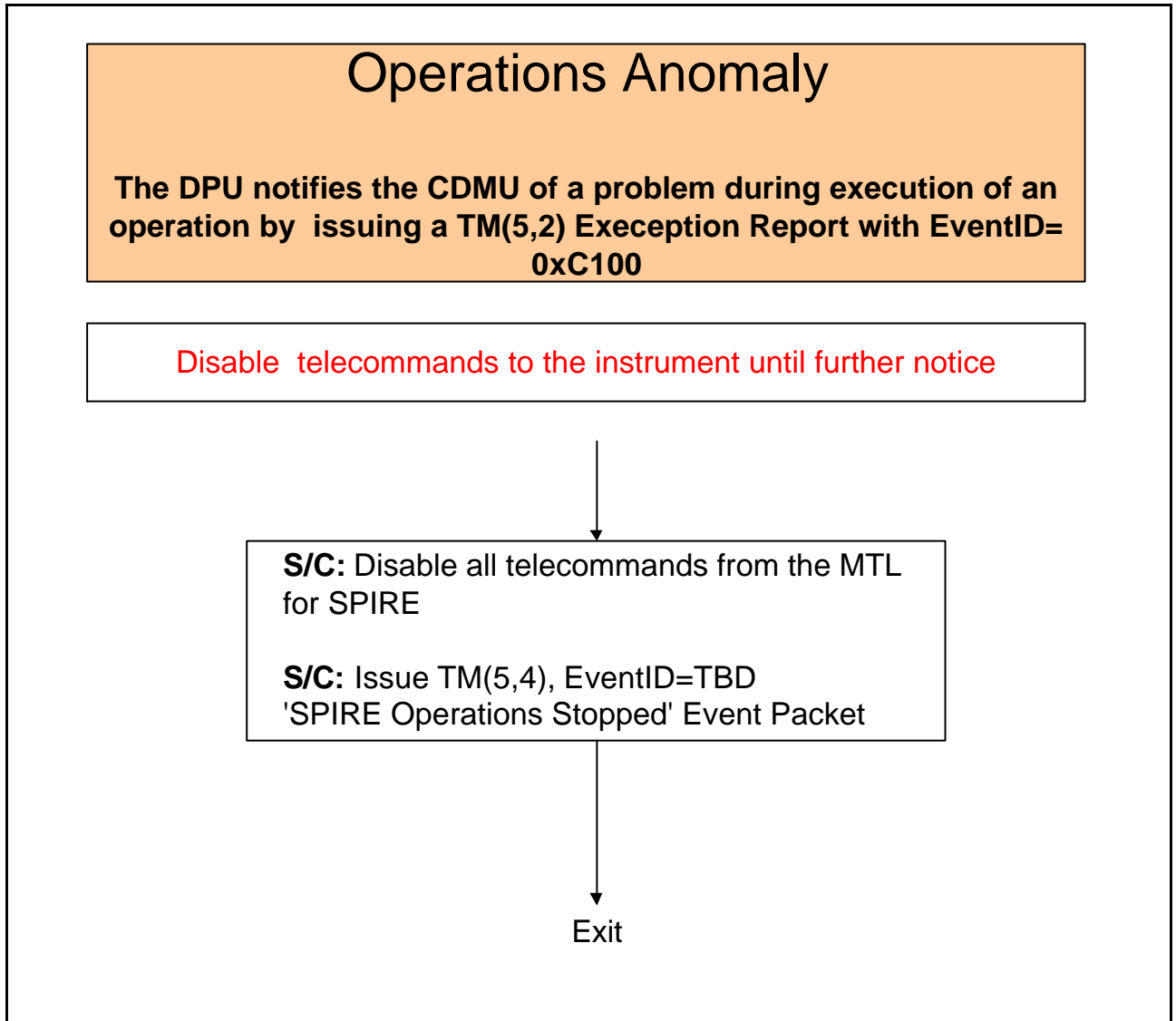


6.3 DPU Anomaly





6.4 Operations Anomaly





6.5 Operations Resume

